

Amendments to the Claims:

1. (Currently amended) In a process for producing a high temperature stable fiber composite ceramic by chemical vapor infiltration (CVI) with a silicon carbide precursor in a suitable carrier gas on fiber scrims of carbon fiber preforms or silicon carbide fiber preforms, the process further comprising ~~the step of~~ adjusting the process pressure to ≥ 0.6 bar absolute and of adjusting the process temperature to $[[\geq]] \geq 1100^{\circ}\text{C}$.

2. (Previously Presented) The process according to claim 1, wherein the process pressure is adjusted to between 0.6 bar absolute and 1.25 bar absolute and the process temperature is adjusted to between 1100°C and 1200°C .

3. (Previously Presented) The process according to claim 1, wherein the silicon carbide precursor is selected from chloro(alkyl)silanes.

4. (Previously Presented) The process according to claim 1, wherein methyltrichlorosilane (MTS) in hydrogen (H_2) as carrier gas is used for silicon carbide deposition.

5. (Currently Amended) The process according to claim 1, wherein a partial pressure ratio of hydrogen to methyltrichlorosilane (H_2/MTS) is adjusted to be between 4 and 8 ~~is adjusted~~.

6. (Currently Amended) The process according to claim 1, wherein a heat-resistant material with a large surface is arranged between ~~[[the]]~~ a gas feed in the reaction space and ~~[[the]]~~ between the fiber scrims of carbon fiber preforms or silicon carbide fiber preforms to be infiltrated for preconditioning the process gas.

7. (Previously Presented) The process according to claim 6, wherein the heat-resistant material with a large surface is a carbon fiber felt.

8. (Previously Presented) The process according to claim 1, wherein carbon fiber scrims are used.

9. (Currently Amended) The process according to claim 1, wherein a residual porosity ~~of is adjusted to be between~~ from 12 to 14 percent of volume ~~percent is adjusted~~.

10. (Previously Presented) The process according to claim 1, wherein the carbon fiber preforms or silicon carbide fiber preforms are generated in that fiber layers are first constructed, the fiber layers are fixed one above the other at a distance from one another by binders, possibly accompanied by simultaneous molding and stabilization of the preform approximating the desired end product.

11. (Currently Amended) The process according to claim 10, wherein the fiber layers are generated with a ~~[[0°/90°]]~~ 0° or 90° laying angle to a main axis of the preform.

12. (Previously Presented) The process according to claim 10, wherein the binder is selected from organic and/or silicon-organic polymer resins.

13. (Previously Presented) Structural component parts, particularly for aircraft and spacecraft, produced according to the process of claim 1.

14. (Previously Presented) Structural component parts for control flaps, leading edges of blades, nose cones, control rudders or heat protection paneling for orbital gliders and hypersonic aircraft produced according to the process of claim 1.